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orthotropic
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Example:
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Moment of Inertia
of a Standard
Structural Shape
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fatigue failure
modes can occur:
fiber breakage,
matrix cracking
and fiber-matrix
debonding. Fiber
Breakage has a
strong dependence
on fiber strength.
Matrix cracking can
be postponed if the
fibers have high
stiffness which

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limits the
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Fatigue analysis of textile composites:
(a) flow chart of the calculation; (b) stages A, B, C of fatigue cycle jump; (c) redistribution of stresses after fatigue damage.

The outlined

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Smith
approach was applied to cross-ply UD laminates (Lian and Yao, 2010 , Shokrieh and Lessard, 2000) and to woven composites (Hanaki et al., 2007 , Kari et al., 2008).

~~Modelling high-cycle fatigue of textile composites~~

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From a certain point of view, fatigue in composites and fatigue in metals are similar: both begin with damage initiation, followed by damage propagation, and end in ultimate failure. Fatigue life N_f , or number of

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cycles to failure,
for both can be
thought of as the
sum of the cycles
during damage
initiation N_i and
the cycles during
damage
propagation N_p .

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The fatigue failure mode of composite materials is an explicit function of the angle between the fiber direction and the applied load; higher fatigue stresses are attributed to smaller angles.

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modes can occur:

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fiber breakage, matrix cracking and fiber-matrix debonding. Fiber Breakage has a strong dependence on fiber strength. Matrix cracking can be postponed if the fibers have high stiffness which limits the

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Maximum tensile stress-N fatigue curves of the investigated composites are reported in Figure 2(a). It is immediately evident the

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relatively low fatigue sensitivity of CF200 laminate, while the higher tensile strength of BF200 with respect to GF200 is responsible for the better fatigue performance of the basalt laminate.

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Fatigue of Textile Composites provides a current, state-of-art review on recent investigations on the fatigue behavior of composite materials, mainly those reinforced with textiles. As

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this particular group of composite materials is extremely important for a wide variety of industrial applications, including automotive, aeronautical, and marine, etc., mainly due to their peculiarities and

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advantages with respect to unidirectional laminated composites, the text presents comprehensive information on the huge variety of interlacement geometric architectures that are suitable for a broad range of

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different applications, their excellent drapability and versatility, which is highly important for complex double-curvature shape components and three-dimensional woven fabrics without plane reinforcement, and their main

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which are currently

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broad range of
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applications

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damage and failure
of composite
materials is critical
for reliable and
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Bringing together materials mechanics and modeling, this book provides a complete guide to damage, fatigue and failure of composite materials. Early chapters focus on the underlying principles

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governing
composite damage,
reviewing basic
equations and
mechanics theory,
before describing
mechanisms of
damage such as
cracking, breakage
and buckling. In
subsequent
chapters, the
physical
mechanisms

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underlying the formation and progression of damage under mechanical loads are described with ample experimental data, and micro- and macro-level damage models are combined. Finally, fatigue of composite

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materials is discussed using
fatigue-life diagrams. While
there is a special emphasis on
polymer matrix composites, metal
and ceramic matrix composites are
also described. Outlining methods
for more reliable design of

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structures, this is a valuable resource for engineers and materials scientists in industry and academia.

An engineering
approach to predict

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the fatigue life and progressive failure of multilayered composite and textile laminates is presented.

Analytical models which account for matrix cracking, statistical fiber failures and nonlinear stress-strain behavior have been

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developed for both composites and textiles. The analysis method is based on a combined micromechanics, fracture mechanics and failure statistics analysis. Experimentally derived empirical coefficients are used to account for

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the interface of
fiber and matrix,
fiber strength, and
fiber-matrix
stiffness
reductions. Similar
approaches were
applied to textiles
using Repeating
Unit Cells. In
composite fatigue
analysis, Walker's
equation is applied
for matrix fatigue

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Cracking and
Heywood's
formulation is used
for fiber strength
fatigue
degradation. The
analysis has been
compared with
experiment with
good agreement.
Comparisons were
made with
Graphite-Epoxy,
C/SiC and

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composite materials. For textile materials, comparisons were made with triaxial braided and plain weave materials under biaxial or uniaxial tension. Fatigue predictions were compared with test data obtained from plain

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weave C/SiC
materials tested at
AS&M. Computer
codes were
developed to
perform the
analysis.

Composite
Progressive Failure
Analysis for
Laminates is
contained in the
code CPFail.

Micromechanics

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Composites Textile
Composites is
contained in the
code MicroTex.

Both codes were
adapted to run as
subroutines for the
finite element code
ABAQUS and CPFail-
ABAQUS and
MicroTex-ABAQUS.
Graphic user
interface (GUI) was
developed to

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connect CPFail and
MicroTex with
ABAQUS. Xue,
David Y. and Shi,
Yucheng and
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The residual
strength of the

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Composite laminates can be predicted using an analytical model. Both the power law and the wearout models appear to be useful in predicting the fatigue life of the composite laminates. However, because of the slope

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parameter, the wearout model appears to have a slight edge over the power law model, particularly at low fatigue life and higher applied stress. The strength degradation due to cyclic loading in notched laminates was found to be

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extremely small up
to a million cycles.

The residual strength of the fatigue-damaged laminates was found to increase (in proportion to the applied maximum stress with $R = 0.1$) after a million fatigue cycles. Impact loading followed by

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Cyclic loading was found to be more damaging (in reducing the life of the laminate) than the reversed sequence of loading. The magnitude of the minimum projectile velocity causing catastrophic failure in the laminates tested was found

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as a function of the
applied stress and
the number of
fatigue cycles.

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